

PŘÍKAZ K ZACHOVÁNÍ LETOVÉ ZPŮSOBILOSTI

CAA-AD-4-094/98

Datum vydání: 14. prosince 1998

LETADLO - KONSTRUKCE - KONTROLA

Týká se: letadel Airbus A310 všech verzí.

Datum účinnosti: 28. ledna 1999

Provést v termínech: jak je popsáno v DGAC AD 92-106-132(B) R5 (příloha tohoto PZZ).

Postup provedených prací: dle DGAC AD 92-106-132(B) R5.

Poznámky: Provedení tohoto PZZ musí být zapsáno do letadlové knihy. Případné dotazy týkající se tohoto PZZ adresujte na ÚCL technický inspektorát - Ing. Toman. Pokud to vyžaduje povaha tohoto PZZ musí být zapracován do příslušné části dokumentace pro obsluhu, údržbu a opravy letadla. Tento PZZ byl vypracován na základě DGAC AD 92-106-132(B) R5, které nahrazuje 92-106-132(B) R4.

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GSAC

AIRWORTHINESS DIRECTIVE

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Inspection and/or modifications described below are mandatory. No person may operate a product to which this Airworthiness Directive applies except in accordance with the requirements of this Airworthiness Directive.

Translation of 'Consigne de Navigabilité' ref. : 92-106-132(B) R5

In case of any difficulty, reference should be made to the French original issue.

AIRBUS INDUSTRIE

A310 Aircraft

Structure fatigue

This Airworthiness Directive applies to AIRBUS INDUSTRIE A310 aircraft, all certified models.

In order to detect or prevent possible damage associated with a phenomenon of structural fatigue :

1. Perform the inspections and/or modifications defined in the following AIRBUS INDUSTRIE Service Bulletins, in accordance with the thresholds indicated in each of these Service Bulletins or within 1000 flights following the effective date of this Airworthiness Directive revision 3 (June 17, 1995), whichever occurs later.

1.1. S.B. A310-53-2014 Revision 5 and Change Notice 5B :

Fuselage - Center Section - Reinforce bottom joint angle fitting at FR40.

1.2. S.B. A310-53-2016 Revision 5 :

Fuselage - Center Section - FR47 and FR54 area - Provide local reinforcement.

1.3. S.B. A310-53-2054 Revision 2 :

Fuselage - Center Section - Inspect frame reinforcement angle runout of FR46 between STGR21 and 22 LH/RH.

1.4. S.B. A310-53-2057 Revision 1:

Fuselage - Center section - Inspection of the T-section connecting FR50A to the beam between FR50 and FR51 LH/RH.

1.5. S.B. A310-53-2059 Revision 1:

Fuselage - Inspect the lower side panel at the lap joint with the upper side panel at FR47 and STGR22, LH/RH.

1.6. S.B. A310-55-2002 Revision 4 :

Stabilizers - Horizontal stabilizer structure - Reinforcement of junction parts.

1.7. S.B. A310-55-2004 Revision 3 :

Stabilizers - Horizontal stabilizer structure - Inspection program after reinforcement as per Service Bulletin no A310-55-2002 embodiment.

1.8. S.B. A310-57-2002 Revision 2 :

Wings - Leading edge access panels landing - Lower skin - Inspection for cracks at bolt holes.

1.9. S.B. A310-57-2006 Revision 3 :

Wings - Holes around overwing refueling aperture at RIBS 13 -14, Inspection.

1.10 S.B. A310-57-2032 Revision 3 :

Wings - Upper skin forward of front spar - Inspection for cracks.

1.11 S.B. A310-57-2037 Revision 3 :

Wings - Bottom skin aft of rear spar inspection for cracks forward of the ailerons.

1.12 S.B. A310-57-2039 :

Wings - Center wing - Inspection of front spar web vertical posts.

1.13 S.B. A310-57-2046 Revision 6 :

Wings - Rear spar at selected bolt locations for attachment of MLG forward pick up fitting.

1.14 S.B. A310-57-2047' Revision 3 :

Wings - Center wing - Inspect rear spar internal angle and tee fitting.

1.15 S.B. A310-57-2050 and Change Notice OB :

Wings - Center wing box - Inspect treillis boom drain holes.

1.16 S.B. A310-53-2074 Revision 2 :

Fuselage - Tail cone - Inspect the lower horizontal-stabilizer cutout longeron, the corner fitting, the skin stap and the skin between FR87 and FR89 and between STGR24 and STGR27, LH and RH.

1.17 S.B. A310-57-2064 :

Fuselage - FR40 - Inspect upper corner fitting.

1.18 S.B. A310-57-2038 Revision 3 :

Wings - Stringer flanges at rib 14 wing bottom skin - Inspect for cracks.

NOTE 1:For paragraphs 1.1, 1.10, 1.11, 1.17, 1.18 comply with the instructions of above-mentioned Service Bulletins at the defined thresholds or within 1000 flight following the effective date of this Airworthiness Directive Revision 4, whichever occurs later.

2. Repeat the inspections defined in the Service Bulletins mentioned in paragraph 1/ above, in accordance with the intervals indicated in each of the Service Bulletins when applicable.

NOTE 2 : The thresholds and intervals which are established and valid for aircraft operated for an individual average airborne flight time of 96 minutes must be adjusted in accordance with the instructions given in the Service Bulletins quoted above.

Ref. : Following AIRBUS INDUSTRIE Service Bulletins, at the latest issue :

A310-53-2014, -2016, -2054, -2057, -2059, -2074,

A310-55-2002, -2004

A310-57-2002, -2006, -2032, -2037, -2038, -2039,

-2046, -2047, -2050, -2064.

This Revision 5 replaces Airworthiness Directive 92-106-132(B) R4 dated June 05, 1996.

Effective dates:

Original AD: May 09, 1992

Revision 1: May 09, 1992

Revision 2: January 28, 1995

Revision 3: June 17, 1995

Revision 4: June 15, 1996

Revision 5: November 28, 1998